

IN THE SPECIFICATION

Page 13, the second full paragraph, lines 18 to 29,  
replace the paragraph with:

On the other hand, in a condition where the heat resisting pipe 70 is completely inserted into the first stage turbine disk 11, the front end portion contacts with the projecting step portion ~~71~~ 81, and in conjunction therewith, the ring shaped projecting portion 71 is received within the spot facing recess 76 with tightly engaging therewith. Furthermore, in a condition where the spacer disk 15 between the first stage and the second stage is stacked on the first stage turbine disk 11, the ring shaped projecting portion 71 is arranged in opposition to the front stacking plane of the spacer disk 15 between the first stage and the second stage in proximity thereof.

On page 14, the second full paragraph, lines 26 to 31,  
replace the paragraph with:

The ~~disc~~ distant piece 3 is coupled with stacking on the front stacking place of the first stage turbine disc 11, and is connected with a not shown compressor rotor in further front side. On the other hand, on the rear stacking plane, a slit 41 communicated with the coolant supply path 7 of the first turbine disc 11 extends toward the outer periphery.

On page 18, the first full paragraph, lines 2 to 17,  
replace the paragraph with:

The coolant ~~62~~ 61 cooled the first stage blade 21 is  
introduced into a cavity 32 formed on the outer periphery side

between the first turbine disc 11 and the spacer disc 15 between the first stage and second stage through discharge ports 27 formed in the dovetail 25 of the first stage blade 21 and collection holes 52 of the first stage turbine disc 11. The cavity 32 and the coolant recovery path 8 are communicated through slits 43 formed on the front stacking plane of the spacer disc 15 between the first stage and second stage. The coolant 62 after cooling the blades flows into the coolant recovery paths 8 from the cavity 32 through the slits 43. The coolant 62 passed through the coolant recovery paths 8 is discharged from the coolant recovery port 6 via the slits 45 formed on the front stacking plane of the stub shaft 2 and through the collecting passages 10 in the stub shaft formed in the axial center portion in the stub shaft 2.

On page 18, the second full paragraph, lines 18 to 27, replace the paragraph with:

On the other hand, similarly, the coolant-~~62~~ 61 cooled the second stage blade 22 is introduced into a cavity 33 formed on outer periphery side between the spacer disc 15 between the first stage and second stage and the second stage turbine disc 12 via the discharge ports 28 in the dovetail of the second stage blade 22. The coolant in the cavity 33 flows into the coolant recovery path 8 via slits 44 formed on the rear stacking plane of the spacer disc 15 between the first stage and second stage and is discharged from the coolant recovery port 6 via the stub shaft 2.

On page 24, the third full paragraph, lines 16 to 19, replace the paragraph with:

Fig. 7 is an enlarged illustration of the portion C in Fig. 1. Fig.-6\_7 shows an alternative embodiment, in which the annular seal member of O-shaped (hollow circular shaped) cross-section is used.

On page 24, the fifth full paragraph, lines 28 to 31, replace the paragraph with:

Fig. 8 is an enlarged illustration of the portion C in Fig. 1. Fig.-6\_8 shows an alternative embodiment, in which the annular seal member of C-shaped (hollow circular shaped) cross-section is used.